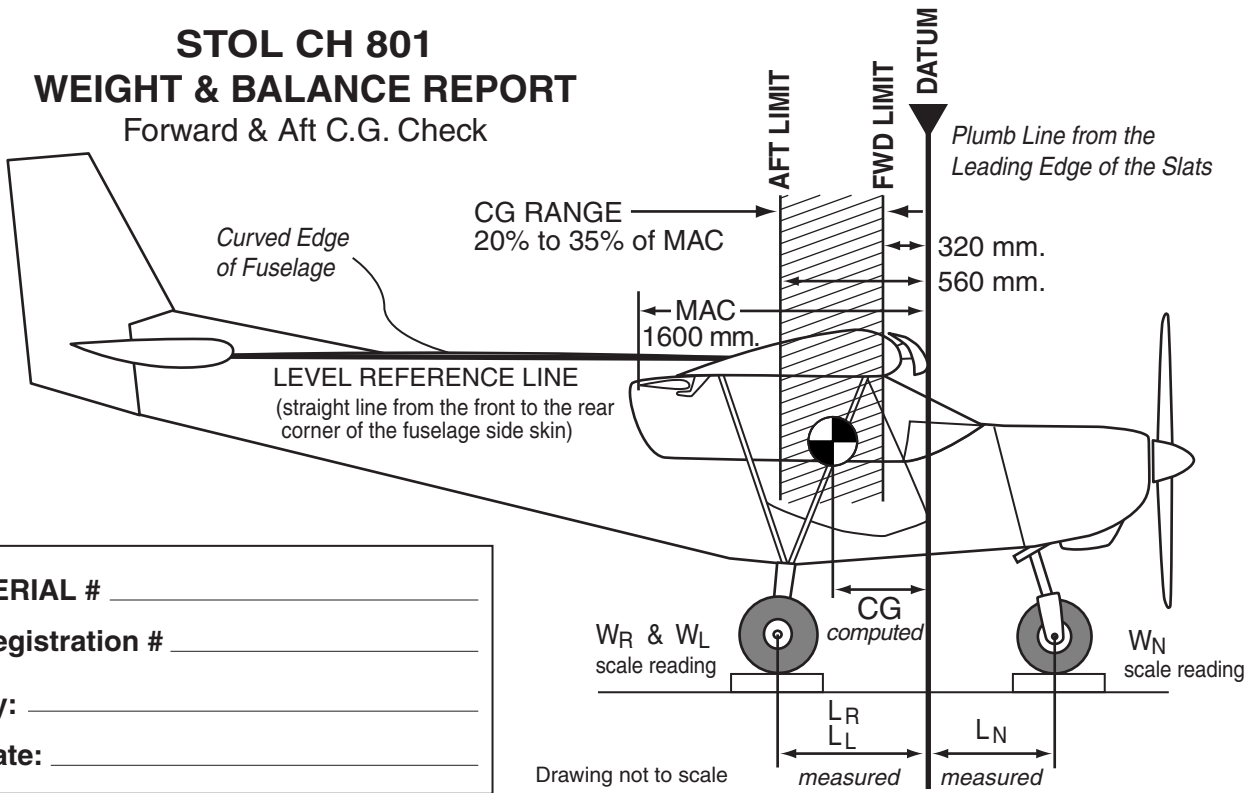


# STOL CH 801 WEIGHT & BALANCE REPORT

Forward & Aft C.G. Check



**SERIAL #** \_\_\_\_\_

**Registration #** \_\_\_\_\_

**By:** \_\_\_\_\_

**Date:** \_\_\_\_\_

Drawing not to scale

ITEM	WEIGHT (pounds)	ARM (mm.)	MOMENT
<b>AIRCRAFT EMPTY CG</b>	RIGHT MAIN WHEEL	$W_R =$	$L_R =$ $W_R \times L_R$
	LEFT MAIN WHEEL	$W_L =$	$L_L =$
	NOSE WHEEL	$W_N =$	$L_N = -$ <i>negative arm</i>
	COMPUTED CG EMPTY	Empty Weight:	$CG =$ <i>Arm to Datum</i>

		MOMENT - Forward	MOMENT - Rear
PILOT		500 *	
PASSENGER		500 *	
REAR SEAT		1,400	
FUEL: WING TANKS L & R: ____ Gal. each		800	
FUEL: EXTENDED RANGE		800	
<b>TOTAL</b>	$W_F =$	$M_F =$	$M_R =$
	$W_R =$		
Gross Weight:	Take-Off Weight:	$CG_{FRD} =$	$CG_{AFT} =$

\* Depends on seat position & individuals' weight distribution.

CG Range: From 320 mm. to 560 mm.

Center of Gravity (CG) =  $\frac{\text{Total Moment}}{\text{Total Weight}}$